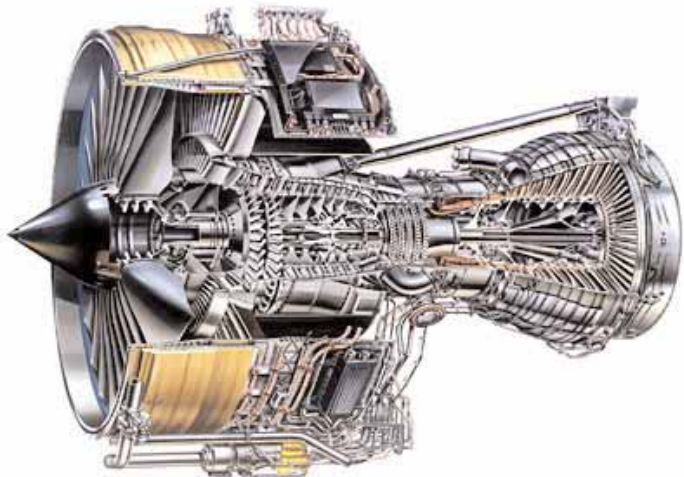




**von KARMAN INSTITUTE
FOR FLUID DYNAMICS**

**AEROENGINE DESIGN:
FROM STATE OF THE ART
TURBOFANS TOWARDS
INNOVATIVE ARCHITECTURES**



March 3-7, 2008

	von Karman Institute for Fluid Dynamics 72, Chaussée de Waterloo 1640 Rhode-Saint-Genèse, Belgium
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INTRODUCTION

The first lectures will focus on the **preliminary design** of state of the art turbofan engine for the propulsion of civil aircraft. Issues like thermodynamic cycle, mission analysis, and off design operation (operating line) will be addressed. Practical examples will be presented. Once this important phase is completed, the different design teams have the boundary conditions to start the design of each component.

This starts with the **fan** that must cope with a number of constraints in terms of mechanical resistance, noise and surge margin. The aerodynamic design involves both subsonic flow at the hub and supersonic flow at the tip. The design of the **booster** (low pressure compressor) will then be addressed where the multistage operation and the matching between successive stages are important concerns.

Moving to the **high pressure compressor**, the lectures will first describe the preliminary design based on throughflow calculations. Then, airfoil design, from 2D sections to stage matching optimisation will be addressed. Current trends for future HPC configurations will be outlined.

The design of the **combustion chamber** involves a number of disciplines such as aerodynamics, fuel atomisation, chemistry of combustion and combustor cooling as well as environmental regulatory issues for emissions (NOx, CO, UHC, soot) along with future combustor technologies. Each of these topics will be reviewed while addressing the design of the combustion chamber.

The **high-pressure turbine** has the particularity to be submitted to high levels of centrifugal force and exposure to very hot burned gases. The design process, involving successively 1D, 2D and 3D analysis is targeting the highest stage efficiency while accommodating for the cooling of the blades.

The **low-pressure turbine** design must satisfy high efficiency together with low weight, cost and noise. This is defined primarily by the load and flow coefficients and the number of airfoils. New lightweight/lowcost configurations tend to reduce the number of blades thanks to high lift designs that take advantage of the positive effects of the unsteady row interaction on the airfoil boundary layer behaviour.

Concerns related to fuel consumption and environmentally friendliness have pushed the engine manufacturer community to look into innovative architectures. Some of them are under investigation in the large European projects VITAL, NEWAC, and soon in DREAM. One possibility is to improve the **heat management** in the engine taking advantage of the available hot (turbine exhaust) and cold (external air) heat sources. This leads to an increase of the thermodynamic efficiency of the engine cycle.

Another alternative is to revise the approach to the main source of thrust and noise i.e. the fan. The **geared turbofan** provides lower fan tip speeds with a high speed booster and a high speed low pressure turbine opening perspectives for a more silent, compact and lighter engine. The **2 stage contra-rotating fan** targets similar objectives with a lower fan tip speed, a more compact nacelle (less drag) and a lighter engine. Finally the **open rotor** configuration opens an alternative for a significant reduction of the fuel consumption thanks to an improved propulsive efficiency.

The Lecture Series directors are Dr. R. Dénos, DG Research, 'Aeronautics' and Prof. G. Paniagua, Turbomachinery and Propulsion Department, von Karman Institute.

TIMETABLE

MONDAY MARCH 3, 2008

- 09:15 Welcome address**
M. Carbonaro, von Karman Institute for Fluid Dynamics, Belgium
- 09:30 Preliminary design**
J. Kurzke, GasTurb Software, Germany
- 11:15 Preliminary design (Continued)**
J. Kurzke
- 14:00 Fan**
N. Smith, Rolls Royce plc, United Kingdom
- 15:45 Fan and booster**
N. Smith
- 17:00 Reception**

TUESDAY MARCH 4, 2008

- 09:00 HP compressor**
O. Damerq and T. Obrecht, Snecma, France
- 10:45 HP compressor (Continued)**
O. Damerq and T. Obrecht
- 14:00 Combustion chamber**
T. Doerr, Rolls Royce Deutschland Ltd, CoKG, Germany
- 15:45 Combustion chamber (Continued)**
T. Doerr

WEDNESDAY MARCH 5, 2008

- 09:00 LP turbine**
P. de la Calzada, ITP, Spain
- 10:45 LP turbine**
P. de la Calzada
- 14:00 Laboratory tour**

THURSDAY MARCH 6, 2008

- 09:00 HP turbine**
F. Haselbach, Rolls Royce plc, United Kingdom
- 10:45 HP turbine**
F. Haselbach
- 14:00 Heat management**
G. Wilfert, MTU Aero Engines, Germany
- 15:45 Geared fan**
G. Wilfert

FRIDAY MARCH 7, 2008

- 09:00 Contra rotating fan**
N. Tantot, Snecma, France
- 10:45 Open rotor**
N. Tantot
- 14:00 VKI bus departure**

PRACTICAL INFORMATION

Lunch will be taken from 12h30 to 14h00. Coffee breaks are scheduled each morning and afternoon.

